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### 论文资料

뮹:

中文摘要:

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题: 热结构与热防护

气动加热对主动冷却超声速燃烧室热结构的影响 中文标题:

英文标题: EFFECT OF AERODYNAMIC HEATING ON ACTIVELY COOLED SUPERSONIC COMBSUTOR

> 气动加热对主动冷却系统的换热性能与安全行为有何影响是一个值得关注的问题。本文提出了一种主动冷却燃 烧室内外热环境的耦合分析方法,考虑了燃烧室外部气动加热、内部对流换热以及冷却通道对流换热之间的耦合 关系。针对典型的主动冷却壁板结构,在飞行马赫数5、6、7三种条件下,具体分析了内外流耦合传热及热结构

> 响应,初步获得气动加热对燃烧室结构、冷却通道及燃料的影响规律。通过对油温、结构温度、屈服条件及变形 量等几个方面分析,建立了主动冷却结构安全裕度判据。马赫数6以上,内外流耦合效应使结构和燃料均已进入

危险区,为此提出初步的隔热设计方案。

Aerodynamic heating of the external flow is an important factor for the cooling efficiency and safe utility of an actively cooled supersonic combustor. In this paper, a heat transfer analysis coupling the aerodynamic heating, the convective heat transfer of the combustor internal flow and the heat absorption of the coolant flow was developed. The analysis was applied for the heat analysis of an actively cooled combustor wall with the effect of the external aerodynamic heating at flight Mach number of 5, 6 and 7. The coupled heat transfer process and the thermal/stress properties of the cooled wall were studied and preliminary conclusions about the effect of the

aerodynamic heating were obtained. A criterion for the safety of the cooling structure was proposed based on the analysis of the fuel temperature, structure temperature, yield condition and deformation. It is found that at a flight Mach number higher than 6, the combined effect of the external aerodynamic heating and the internal combustor flow would cause the wall structure and the coolant (hydrocarbon fuel) enter an unsafe region of usage. According to the present results, a primary design of the insulation of the external aerodynamic heating was discussed.

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